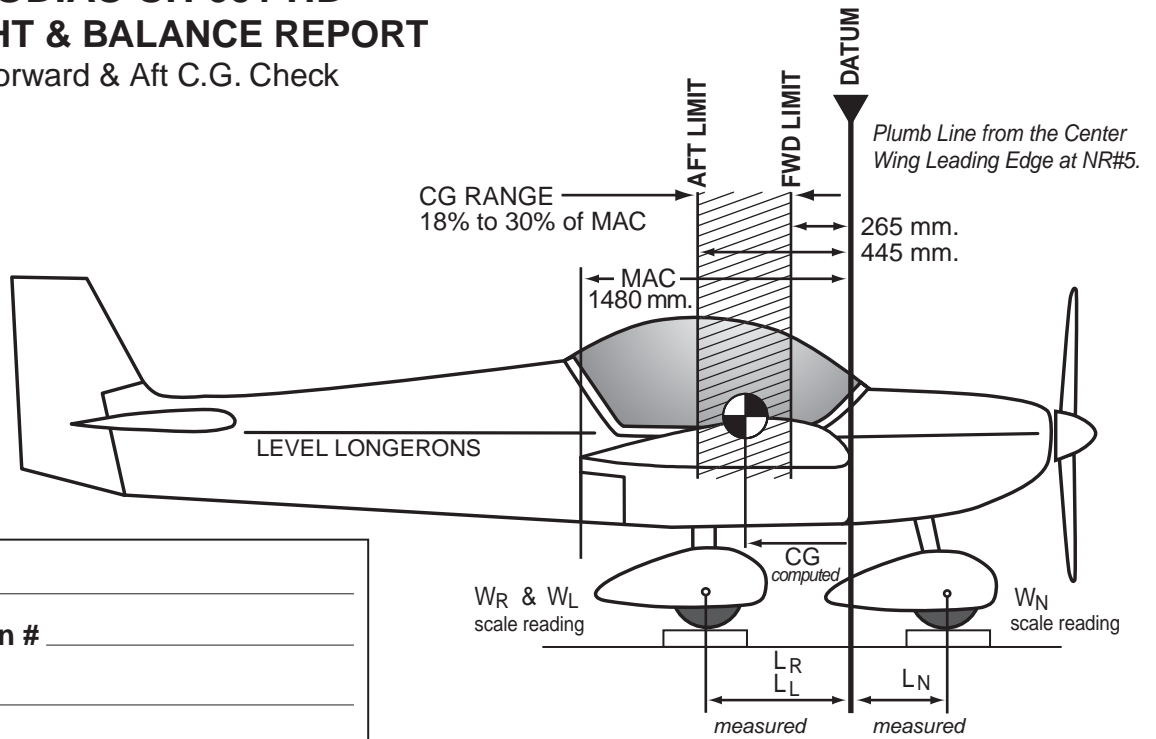


# ZODIAC CH 601 HD

## WEIGHT & BALANCE REPORT

Forward & Aft C.G. Check



Drawing not to scale.

**SERIAL #** \_\_\_\_\_

**Registration #** \_\_\_\_\_

**By:** \_\_\_\_\_

**Date:** \_\_\_\_\_

|                          | ITEM                               | WEIGHT (pounds)  | ARM (mm.)                         | MOMENT                 |              |
|--------------------------|------------------------------------|------------------|-----------------------------------|------------------------|--------------|
| <b>AIRCRAFT EMPTY CG</b> | RIGHT MAIN WHEEL                   | $W_R =$          | $L_R =$                           | $W_R \times L_R$       |              |
|                          | LEFT MAIN WHEEL                    | $W_L =$          | $L_L =$                           |                        |              |
|                          | NOSE WHEEL                         | $W_N =$          | $L_N = -$<br><i>negative arm</i>  | -                      |              |
|                          | COMPUTED CG EMPTY                  | Empty Weight:    | <b>CG=</b><br><i>Arm to Datum</i> | <i>Aircraft Moment</i> |              |
|                          |                                    |                  | <b>MOMENT - Forward</b>           | <b>MOMENT - Rear</b>   |              |
|                          | PILOT                              |                  | <b>650</b>                        |                        |              |
|                          | PASSENGER                          |                  | <b>650</b>                        |                        |              |
|                          | BAGGAGE (rear)                     |                  | <b>1,400</b>                      |                        |              |
|                          | FUEL: Header Tank<br>16 US Gallons |                  | <b>- 100</b>                      |                        |              |
|                          | Wing Baggage Lockers               |                  | <b>600</b>                        |                        |              |
|                          | <b>TOTAL</b>                       | $W_{FRD} =$      |                                   | $M_{FRD} =$            | $M_{AFT} =$  |
|                          |                                    | $W_{AFT} =$      |                                   |                        |              |
|                          | Gross Weight:                      | Take-Off Weight: |                                   | $CG_{FRD} =$           | $CG_{AFT} =$ |

CG Range: From **265 mm.** to **445 mm.**

$$\text{Center of Gravity (CG)} = \frac{\text{Total Moment}}{\text{Total Weight}}$$